

METHODS AND APPARATUS FOR COOLING GAS
TURBINE ENGINE ROTOR ASSEMBLIES

ABSTRACT OF THE DISCLOSURE

A method facilitates fabricating a rotor assembly for a gas turbine engine. The method comprises providing a plurality of rotor blades that each include an airfoil, a dovetail, a shank, and a platform, wherein the platform extends between the shank and the airfoil, and wherein the dovetail extends outwardly from the shank, and forming a cooling circuit within a portion of the shank to supply cooling air to the rotor blade for supplying cooling air to the rotor blade for impingement cooling a portion of the rotor blade and for supplying cooling air to the rotor blade for purging a cavity defined downstream from the rotor blade.